

Amendments to the Specification:

Please replace paragraph [0025] with the following amended paragraph:

[0025] Broadly, the invention provides an innovative cooling tube for reducing the temperature of an object, such as the liner of a combustor in a gas turbine engine. The cooling tube of the present invention departs from the aforementioned configuration of the prior art exemplified in FIG.1. The cooling tube 160 provided by the current invention is shown in FIG. 2 on the same curved surface 220 and projected onto plane 260. Cooling tube 160 has the characteristic that no plane exists where the first end 161, second end 162, and the centerline 302 of cooling tube 160 project onto a straight line in the plane. In general, the present invention provides a plurality of bends in the cooling tube 160 along its extent between its two ends 161, 162, such that the centerline 302 of the cooling tube 160 is not in a plane. In this case, the centerline 302, the first end 161, and the second end 162 of cooling tube 160 are non-linear when they are projected onto the plane. The cooling tube 160 can follow a curved, or serpentine, path along the surface 220 from one end to the other. If the surface 220 is a combustor liner, then the cooling tube 160 is exposed to steep temperature gradients between itself and the surface 220 to which it is attached, and the bends in the cooling tube 160 can allow it to compensate for the expansion and contraction that may result from such thermal stress.